FLUENT - Flow over an Airfoil- Problem Specification

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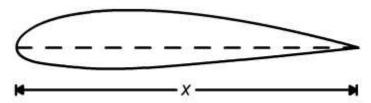
Problem Specification

- 1. Create Geometry in GAMBIT
- 2. Mesh Geometry in GAMBIT
- 3. Specify Boundary Types in GAMBIT
- 4. Set Up Problem in FLUENT
- 5. Solve!
- 6. Analyze Results
- 7. Refine Mesh

Problem 1

Problem 2

Problem Specification



Consider air flowing over NACA 4412 airfoil. The freestream velocity is 50 m/s and the angle of attack is 2°. Assume standard sea-level values for the freestream properties:

Pressure = 101,325 Pa

Density = 1.2250 kg/m3

Temperature = 288.16 K

Kinematic viscosity v = 1.4607e-5 m2/s

We will determine the lift and drag coefficients under these conditions using FLUENT.

Go to Step 1: Create Geometry in GAMBIT

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