Problem Specification

Consider air flowing over NACA 4412 airfoil. The freestream velocity is 50 m/s and the angle of attack is 2°. Assume standard sea-level values for the freestream properties:

- Pressure = 101,325 Pa
- Density = 1.2250 kg/m³
- Temperature = 288.16 K
- Kinematic viscosity $\nu = 1.4607 \times 10^{-5}$ m²/s

We will determine the lift and drag coefficients under these conditions using FLUENT.

Go to Step 1: Create Geometry in GAMBIT

See and rate the complete Learning Module

Go to all FLUENT Learning Modules